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AVIATION

FOR APRIL, 1934

Air Mail Legislation in the Making

Congress debates new codes full of startling features

WHATEVER may come out of the present confusion of committees on the subject of air transport, at least one thing is certain: We are about to see the execution of another landmark change in our mail policy and in the lines of the financial relations between the transport laws and the government. These great landmarks seem to be decided to arrive approximately at three-year intervals. The first was the Kelly Act of 1925, that first made it possible for commercial airline operators to get some kind of financial encouragement from the government. The second major modification was the re-issuance of the act of 1928, recognizing the principle that there must be a substantial guarantee of continued government assistance over a much longer term than that of a four-year contract of property equipped and properly equipped transport systems were to be developed. Next came the Warren Act of 1930 changing the system of compensation for carrying the mail and giving the Postmaster General a degree of freedom that enabled him to encourage and to compensate provision for carrying passengers on mail lines and allowed him also to encourage, among possible agencies with a view to concentrating the rights of powers in the field and to give well-organized enterprises, backed by personnel of demonstrated experience and competence, a preference over non-line operations. Now it is 1934, and new legislation based on the practices in-

concluded since 1930 is perhaps already overdue. Five more than a year ago plans have been in the making, the subject of pending study by the Post Office Committee of the House of Representatives but with the dramatic gesture of contract expiration on Feb. 10 the time for pending deliberation came to an end. Manifestly if we are to have any air mail service at all, or any other kind of air transport, we have to get some new legislation without further delay. Both houses of the Congress have been forced to start a high-priority re-consideration of the subject.

A series of bills

From Feb. 10 to March 10 there have been introduced some 40 air mail bills. Most of them have aimed at such things as renewal of a renewable bill, as the ground in the last session, the Kelly bill of 1925, has expired in a three-year term, with an easy different Congressmen's names attached and embodying every conceivable combination of possible rates of payment upon every sort of a sliding scale. From the whole also there emerged steel and one instance of intercontinent experience, notwithstanding because it represented the apparent present desires of the Executive as recorded in his letter of March 7 to the chairman of the respective committees of the Senate and the House of Representatives. Within 48 hours after receipt of the letter drafted bills had been introduced into the two houses (in-

cluding that hearings might start in both places at once). The House bill has the name of Mr. Mand of Buffalo, N. Y., chairman of the Committee on Post Offices and Post Roads. The Senate version, drafted in fact, is remarkable in that it appears under joint sponsorship, the credit for its appearance after two weeks between Senator McKellar of Tennessee and Senator Black of Alabama. Senator McKellar is chairman of the Senate committee dealing with postal affairs, and although he was one of the warm advocates of the Warren Act at the time of its passage and showed a particular interest in the possibility that the Postmaster General's privilege of carrying fees might be used to create a new air mail route through his (Senator McKellar's) state he has since become a pronounced skeptic on the necessity of maintaining air mail operations at anything like the increased level. Within the past few weeks he has been one of the most energetic upholders of the Postmaster General's office in cancelling the war mail routes, and has had an opportunity of expressing his conviction that fiscal and collection had been simply proved and that as further discussion on these points was in order. His collapse in the sponsorship of the new legislation, Senator Black, needs no explanation in anyone interested in aviation. As chairman of the special committee that has been engaged in an extensive search for the appropriate air mail in American air transport over a

This is the second of two articles on the 1933 survey flight of Col. and Mrs. Charles A. Lindbergh, written by Commander Weems and based on information furnished by Colonel and Mrs. Lindbergh. As in the first article, Commander Weems has also drawn on such sources as Pan American Airways, of which Colonel Lindbergh is technical advisor, and the American Museum of Natural History, to which the airplane and its equipment has been presented. This concluding installment is devoted largely to a discussion of the long over-water sections of the flight and the equipment used in celestial navigation.

The Flight of The "Tingmissartok"

The authoritative account of the 1933 survey flights of the Lindberghs

By Lt.-Commander P. V. H. Weems
U.S.N. Retired

ON THE long over-water flight from Montreal to Natal, Dec. 8, 1933, full use was made of the radio, including radio, and celestial navigation. Group problems were met and successfully solved. British Admiralty Mercator Charts of the South Atlantic coast (4241, 1 727/30) were used on this leg, Chart 2063A for the eastern section, and Chart 2063 for the western section. It was found that a straight rhumb line or two chords in low latitudes was sufficiently close to the great circle for practical purposes. As noted, the plotted course was divided and marked for 100-mile intervals. The use and magnetic courses were written directly on the charts.

The charts in a conventional Mercator scale showed the variation for every two degrees, as well as islands and the contours of the shores of Africa and South America. Lines of position obtained by radio or by celestial navigation, were plotted directly on the chart.

Wind veers unaccounted

The latest weather charts indicated a southeast wind over Africa, and a westerly wind over South America, with the force and direction of the wind in each case shortly equal and fairly constant. Since the weather was about normal, Colonel Lindbergh estimated that the wind drift on the first half of the leg would be compensated by the opposite wind on the last half. It turned out to be about that way, though of course he does not remember this as a fixed rule.

The compass course was therefore set allowing for the variation and the deviation, but making no allowance for the wind. The first sextant observation showed the plane to be about 40 miles off the route, but an observation was made to get back on the course. Later, another series of observations were taken near the island of St. Paul's Rocks, which was not far from the destination. The position from these observations showed the plane to be nearly on the original course. At this time, however, Mrs. Lindbergh put a radio bearing from the S. S. Windward. The plane's position had been definitely determined by celestial navigation and the strength of the signal indicated the Windward was near the plane on the port beam. It was therefore decided to turn sharp left and follow the radio bearing until the Windward was sighted. After flying along 40 miles on a westerly course, the ship was sighted, after which a new and direct course was set for Natal.

Radio and celestial navigation

Colonel Lindbergh states that on these long flights a sextant is essential. Also, by using lines of position obtained by radio as well as those obtained by celestial navigation, he definitely places three mile zones of position before every water in the category of practical air navigation. In fact, this has been known and perhaps used to a small extent, not in the knowledge of the writer, his flight is the best example of the use of distance of observation. In fact, it was found that greater ac-

curacy in the use of the sextant was possible when the plane was steered by the directional gyro than when it was steered by the magnetic compass. It is possible to hold the plane much steadier in the former case.

Teams work on the flight

The many duties in flight were about evenly divided. The principal tasks were taking the plane, operating the radio, taking and reducing celestial navigation observations, and keeping the log of the plane. These duties were divided fairly evenly between the other two qualified aviators—first-aid officer of Wiley Post and Harold Gatty. Post did software for the plane, and Gatty did software for navigation. In fact the *Winnie Mae* was not just for dual control and communication between Post and Gatty, but was done by voice tubes and in a certain sense on a string. Furthermore, file stations was given to, and practical benefit derived from the use of radio, and therefore some work was devoted to dead reckoning and celestial navigation. As a matter of fact, part of the difference between the two flights may be accounted for by the different seasons of each.

The Transatlantic was fixed with dual controls, the two seaplanes were co-ordinated, and both the Colonel and Mrs. Lindbergh were capable of doing any of the work involved. Actually, however, it was logical to make a clear-cut division of duties, and this was done. Colonel Lindbergh did most of the plot-

ting, and was relieved at the controls whenever he was to take sextant observations and to reduce the observations to positions, and in some cases to make notes and to work on the charts. Von Lindbergh did all the radio sending and receiving work, made all direct and ground speed observations, kept the official log of the flight, and handled the controls when the Colonel was otherwise occupied.

Some details of navigation

The accompanying chart shows some of the navigations accomplished on the *Tingmissartok* flight. No radio methods or "guides" were used on the ocean flight. The navigation was not based on standard (two-principal) methods were used several celestial observations were made for each leg of position "fixes." A sextant was utilized to avoid the usual reliance on observations for getting the strength of the sun or more observations. Adding several times and the corresponding altitudes in air, then dividing them to obtain the average, is an accurate method for obtaining the desired course part of the calculations. Colonel Lindbergh assumed that if he could get as

many as three consecutive altitudes taken at equal intervals of time, that the results of one of the three would not be far off. It was also found that sextants of about the same make were accurate and practically interchangeable. In smooth air it was comparatively easy to get three consecutive satisfactory observations, but in heavy air more observations had to be taken in order to get a string of three good sextant altitudes. When finally settled into the observation track, one of the three sights was worked out, the line of position plotted, and necessary changes to the navigation made. Finally, as a check, one of the remaining sights was worked independently, though the resulting line of position was not plotted. It is not sure it found necessary to allow points about making celestial observations. It was not really essential to alter course because of drift and ground speed observations, so an outline of the flight was known well in advance to an accurate course at the time of take-off, and the compass and directional gyro offered an accurate means for obtaining the desired course.

As Colonel Lindbergh expressed it, one should expect to reach the destina-

tion by straight dead reckoning, while radio and celestial navigation give additional factors of safety, convenience, and a definite check on the navigation as well as on the time. Although any one of the three available methods makes under good conditions, the additional two methods surely provide a considerable factor of safety under most conditions. In staying low nearly the fixed reckoning worked out, Colonel Lindbergh repeatedly obtained the essential need of both the radio and the sextant. A fixed navigational method would require and utilize of three methods.

Long range radio

On the *Buffham* Naval log, Mrs. Lindbergh was in communication with the Pan-American radio stations when her plane was over Africa. Following are samples of the messages received by Pan-American stations at Natal:

05 10 15 1933 Lat. 11° 45' North Long. 17° 15' W. Wind 20-25 mph True. Air 1000 ft. 11 15 30 1933—Position Lat. 11° 45' N. Long. 16° 45' W. Alt 1000 ft. 1 15 45 1933—20° N. 1000 ft. 1 15 55 1933—20° N. 1000 ft.

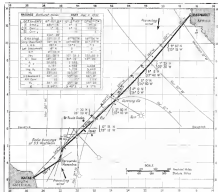
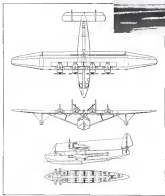


Chart shows the course of the "Tingmissartok" on the South Atlantic coastline. The starting was Montreal and the route was determined from the airplane which is now attached to the plane and is contained in the American Museum of Natural History. (Source: *Natal* in regarding the calculations the difference between the two and the method. The starting course should be 110-110-110.)

A new Super-Clipper for Pan American Airways

More About The Sikorsky S-42



Key: (1) motor wheel; (2) engine; (3) pilot; (4) radio operator; (5) flight instructor; (6) oil tank; baggage and mail pile; (7) forward passenger row; (8) seats behind and above baggage boxes; (9) 30 to 40 horsepower motor; (10) water counterbalance; (11) elevator post and equipment; (12) 30 to 40 horsepower outboard motor (shown with 200-horsepower motor).

OVER two years ago, with the launching of the first of the new Pan-American Clipper ships (see AVIATION, October, 1942), the world's super-seas were taken toward the goal of regular ocean-to-ocean transatlantic service. The S-40s, although a tremendous advance in size and weight over anything previously built in this country, were admitted not the ultimate in trans-Atlantic flying boats but represented an intermediate stage in the evolution of the over-ocean liners of the future. Three years were built and proved prominently successful in plying the Caribbean routes. They in and they out under all sorts of conditions that performance was carefully studied and recorded. When, a year or so ago, the time came to revise specifications for the next step toward the Atlantic service, the lessons learned with the S-40 proved valuable. Thus the S-42 diverges from the history and enters upon her flight track based not only by years of successful design and manufacturing experience by the Sikorsky group of engineers and designers but also by over six years of operating and maintenance experience under the close supervision of Pan American's staff and operating engineers.

Although in performance and carrying capacity the S-42 outranges the S-40 series by a considerable margin, there is no such disparity in dimensions and weight as might be expected. The S-40 and its predecessor the unretreated S-41. The span of the S-42 is 114 ft 2 in. and the gross weight on which the stress analysis of the new ship was based (38,000 lb.) exceeds that of the original Clipper by about 4,000 lb. (The gross weight has been increased 2,306 to 2,700). The weight engine, however, is some 2,008 lb. less than the S-40 (as a flying boat), indicating a great gain in

structural efficiency and consequent improvement in payload. The clue to the better performance lies not only in the added horsepower and the dynamic stress improvements, but also in the smaller total wing area (S-40—3,740 sq ft, S-42—3,338 sq ft.) and the correspondingly higher wing loading. Designer Igor Sikorsky has expressed the opinion that safety and comfort follow from relatively high ratios of weight to wing

area. High loadings mean high loading speed, to be sure, but higher speed also means better control during landing. From a standpoint of safety, present loading speed requirements may have been considerably overestimated.

Although actual performance data are not available at this time of this writing, it is safe to predict that the specifications laid down by Pan American Airways will be met by the S-42. With a



view of five (two pilots, radio operator, mechanic and steward), a full complement of 32 passengers and baggage plus 1,000 lb. of fuel and supplies and a full galley supply (1,500 gal.) the ship is expected to cruise at 150 mph. for a range of some 1,200 miles. With extra loadings and without passengers, but with a considerable payload in fuel and freight, a one-stop range of 2,500 miles can be covered upon making long Trans-Atlantic jumps quite feasible with a moderate margin to spare.

No more bowies

As pointed out in our preliminary discussion (AVIATION, March 1934, page 96) the S-42 differs from other Sikorsky designs in that the sea surfaces are carried directly by the hull and not on open booms. This feature necessitated considerable rearranging of control surfaces and their connection and also made for a somewhat longer hull and more spacious cabin accommodations.

On going aboard the S-42 through the after hatchway, no yank-like preparations are strikingly in evidence. A clean forward ladder brings one over the side and onto level rubber-covered walkways (formed by the open hatch covers) laid back onto the deck. A substantial handrail (shaded in light of course) leads down a solid stairway at the companionway into the cabin. A single side-grip access to all emergency compartments and to the cockpit forward. Bulkhead doors are all well-lit, making it possible to walk off any closed-hatch system in case of damage to the phone. Passenger seats are adjustable and are hung from the bulkheads without legs in the rear. Two completely ruggled seats and work-spaces open off the aisle adjacent to the companionway. Steward's equipment, life rafts, auxiliary fire-fighting equip-

ment and 4 emergency fuel and baggage compartments occupy the space in the bulk between the bulkhead and the wall.

Passenger cabins

Wells and ending to passenger compartments are lined with walnut-bla wood veneer panels. All removable for purposes of underlying structure. Pith of reinforced balsa-foam between the inner panel and the bulkhead panels dampens out vibration and assists in the sound insulation. These circular side doors are provided in each side of each compartment, all fitted with shatter-proof glass and non-opening. Ventilation comes from two overhead ducts with four adjustable registers in each side (shown). Fresh air is taken through a screened opening in the rear of the "lower" construction and wing and delivered through a balanced-type water separator to the ducts at a maximum rate of 42 cu ft. of air per passenger per minute. Since the boats are confined for freight operations, no heating systems have been installed. Control systems are strikingly in evidence. A clean forward ladder brings one over the side and onto level rubber-covered walkways. Floor panels are also removable in small units to give access to the hull lining. The cross-members under the floor between the bulkheads have been spaced for strength so that a man may go down into the bottom without difficulty, and struts are arranged so that no water pipes can come between bulkheads.

Pilot's compartment

The chief pilot is on the left and co-pilot on the right in the usual manner. The radio operator and his apparatus occupy the space behind the co-pilot and the flight instructor is stationed behind the pilot. An interesting admission of a

structure and of accessories has been worked out. Only the essential navigational and engine control instruments (tachometers and manifold pressure gauges) are on the pilot's bench. The simplicity of smaller gauges indicating instruments (such as pressure and temperature gauges, fuel gauges, etc.) are mounted on the rear cockpit bulkhead of the mechanic's post. The pilot is light as thus covered and will bring an instrument and the mechanic has the responsibility for the correct operating conditions for all four engines. The net result is a great simplification in instrument installation and a considerably reduced weight of the instruments and cockpit bulkhead. A floor in the forward cockpit bulkhead gives access to the anchor work and is a highway to the forward dock. The rest of the cockpit is comparatively leads to the outside of the hull on the port side. Removable panels in the wall of the passenger give immediate access to the hull of the passenger's motorway tank. On the other side of the passage sliding drawers are provided for two 32 volt storage batteries. Each is charged by a separate generator.

Structural features

The hull is of all-metal construction (Aluminum) because of its ease of fabrication which gives perfectly smooth surfaces outside. All seams are sealed with marine glue and set in fabric

during assembly. Wing flaps are of the same general type of construction as the main panel, being made up of shaped framing covered with chord sheets.

The reception wing of the S-42 is made up in one piece. The outer sec-

tion ribs are to the control mechanism in the cockpit down through the inspection cover.

The ailerons follow standard Sikorsky practice, having rudders and flaps of a special airtail section to control the air-

craft as arranged on a variable side take-off gear. Each nacelle structure is fully streamlined and the engines fitted by full N.A.C.A. cowlings. Flaps are quickly reconvertible for intermediate. Three-bladed controllable pitch Hamilton Standard propellers are fitted. All pitch control mechanisms are operated from the cockpit. Fuel is carried in eight elliptical shaped tanks on the wings between the spars. Two tanks fuel each engine. Oil is carried in a specially tank between each two main tanks. The outboard engine tanks are provided with quick-acting dump valves. The fitting of the tanks is streamlined so that gas may be pumped in at one point in the center section of the wing for all eight tanks. Oil-corders with the necessary air control valves are mounted under each section in the bottom surface of the wing.

The new S-42 exhibits an extraordinary number of interesting and significant structural details which space limitations preclude full discussion. It is hoped that in later issues of AVIATION we will be able to illustrate and describe many of these outstanding features.



Shows the tail section was supported by the use of the tail. For streamlined take-off and landing, the tail boom and main spar are hinged to the fuselage. The tail boom and main spar are hinged to the fuselage. The tail boom and main spar are hinged to the fuselage. The tail boom and main spar are hinged to the fuselage.

tail which carries the four engines is strengthened in the tail and is rectangular in plan. It carries a balanced landing flap along the full length of the trailing edge. The flap can be set in various pre-determined positions for take-off, cruising and landing, and is provided with an ingenious automatic control which sets it automatically depending on the air speed of the ship.

The wingtip sections are strongly tapered with ailerons running practically the entire length. Aileron control is of the cable and pulley type—the cable from each section running forward to the center of the wing and attached to a cable for differential action. Cables from

balanced forces due to stoppage of one or more engines. All sections are duct framed around with fabric. Rudders and elevators are balanced. Submarine equipment is obtained through a conventional throat and air conditioning equipment uses large hand wheels in the cockpit. Control surfaces are all hinged on self-aligning ball bearings and are all equipped with adjustable tabs for follow-up adjustment. Tabs are adjustable from the ground only.

Power Plant Installation

The four Pratt & Whitney Hornet R-1609, 500-hp engines (rated 3,200 r.p.m. 675 hp at 2,800 r.p.m. at 2,150 p.s.i.)

April 1939

April 1939

Wright Field completes flight and static tests on the Boeing 247

Load Test—Full Scale

ALTHOUGH the value of test work on models is beyond all question, the use during test of actual aircraft paper tables to determine the behavior of a monocoque structure, and quite another to take a transport plane out of active service and put it in destruction. So seldom does a laboratory get such an opportunity as was afforded by the assignment of one of the Boeing 247 airplanes by Wright Field for complete flight and static test. Although the outcome of the tests has been previously reported (AVIATION, March, 1934, page 97) certain details of the procedure and results are of sufficient interest to warrant further discussion.

Strength tests on the Boeing 247 were of particular interest since the plane represented a distinct departure from previously accepted types of construction and marked the beginning of the present trend to metal monocoque landing monoplanes. One of the most important features in the investigation was to determine whether or not that type of construction would permit of greater load and faster schedules than the planes had previously been known to undertake. The fact that under static test the loadings of the plane with standard loads up to 52 tons when a maximum of only 30 tons was required was an indication of the potential strength of the design. Significant also is the fact that the plane subjected to test was not taken from current factory production, but was withdrawn from active scheduled flying after over 1,400 hours of passenger carrying. Tests were made on the basis of a gross load of 114,000 lb., with 500 lb. greater than the gross load and gross weight. In view of the results, application has been made for an increase in the permissible gross load up to 121,000 lb.

Before being put into the laboratory the airplane was flown under a wide variety of conditions by a number of pilots and its behavior carefully watched and recorded. It was flown as smooth and rough air, at all speeds between stall and above 500 in mph, with the wheels up and below stall and landing high speed flights with wheels down. No unusual vibration or drop-off of any kind when flying with the engines. When flying with one engine out the functioning of the rudder in an extreme position, a slow rotation of the control surface was obtained by locking back at the rudder from the rest but was not sufficient to take the rudder panel. This vibra-



Wide section of the Boeing 247 fuselage static testing at Wright Field. Load was not only applied at the structure attachment points as the structure was fitted with steel beam supports. As tests started before full failure occurred the maximum load was not 52 tons but 57 tons. The maximum load was not 52 tons but 57 tons. The maximum load was not 52 tons but 57 tons.

tion did not show any tendency to lighten up and stopped as soon as two-engine flight was resumed.

Although flight tests were interrupted in the light of Air Corps requirements for safety and maneuverability, no outstanding loads were reported. The latter said, "The loads were considered a little heavy" under certain conditions but were easily corrected through the use of the trimming tabs. As far as stability was concerned, it was concluded that more the plane has been allowed for one particular condition of power or load in the case of the tabs, it could be flown "handily" and safely.

After flight testing was completed the ship was moved to the laboratory where it first ran on the ground at the standard Air Corps vibration tests. The natural frequencies of all parts of an airplane structure may be determined by applying vibrations of known amplitude and frequency to mechanical means (see AVIATION, January, 1934, page 21). All possible methods of vibrating equipment, test systems were applied and the frequencies of the structure's lack of resonance was determined. The only portion

of the structure which fell outside the limits imposed by the Air Corps, were the wing doors. The reason why the natural period was felt to be a little too close to the normal rate of vibration of the engine at cruising speed is to be entirely satisfactory. Such a condition, however, is avoidable by changing the rubber mounting brackets. Particular attention was made to the vibration period of the main structure of the fuselage, but since all tail surfaces were in place and other approximations made, no structural conditions developed over the range of loading studied.

With the vibration explorations out of the way, the fuselage was tested with one of two sets and put load and failure occurred. All vibration was made between that portion of the fuselage forward of the center of gravity and aft of the center of gravity. For the entire fuselage the load factors developed under tests were far and away in excess of those required by the Department of Commerce. Critical check was made on the fuselage through the application of loads. For load factors up to and including 3.0 about of the C.G. and 2.75 aft of the C.G. the deflection was measured at seven sections.

At loads equivalent to factors of 3.25 and 3.50, the rear portion of the structure began to settle slightly, indicating that the design load had been reached. Primary failure occurred in the bottom of the mid-section of the fuselage approximately 1 ft behind the rear wing spar. All longitudinal stresses and the skin at this point buckled under compressive stresses. At the point of failure the critical load of the C.G. was 90 per cent and aft of the C.G. 62 per cent. To magnify the story, other failures had occurred samples of the material were removed from various portions of the fuselage structure and the specifications for chemical composition and physical properties. In all cases the materials were found to fall well within the specifications for 7075 aluminum. No evidence of corrosion or fatigue developed.

Wing and landing gear tests were not conducted at this time, but will probably be made at the Boeing factory at Seattle. Wing loads were run up to a load of better than 36 tons, the required factors (1.50, 1.75, 2.0, and 2.25) and (forward) of exceeding the Department of Commerce requirements by a considerable margin. Landing gear tests indicated that the plane could withstand an emergency impact of some 30 tons.

Structure of the Boeing 247. The water-tight skin is replaced from inside, a method which is essential to safety and lower the fuselage weight.

In this, the second of two articles devoted to the resolutions reached in the existing studies made in the course of testing the new Douglas Transport, the authors have defined existing conditions and outlined their introductory remarks regarding methods of existing control. Future articles will outline the conventional testing procedure and present a new method of accurately determining existing performance and controlling existing operation. Since the first of three articles appeared the transcontinental record-breaking flight and many intensity records of the Douglas Transport have served as excellent testimonials to these methods.

Cruising Control in Transport Operation

By Edmund T. Allen and W. Bailey Oswald

THE SECOND OF TWO ARTICLES

COMMERCIAL air transport operators have usually set up their own limitations for existing. Those having a long background of equipment-maintenance problems determine their limitations at a point where the demand for increased speed meets the demands for decreased maintenance costs and increased reliability of performance. If the airline has life or no competition, it is not pushed by the government post office to decrease mail transit time, and air traffic laws, lean passenger revenues by increasing speed, the operating conditions will be set so as to favor long life of equipment and slow schedules. An example of this kind is the National Parks Airways whose operating speed is almost very slow but whose engine failures for that period have been virtually nil. The airline which, on the other hand, is competing stiffly on a speed basis with other lines, such as the Varney Speed Lines, will set its limitations so as to favor high existing speed and occasionally greater maintenance costs.

In either case, whether set up for high existing speed schedules or for slow ones, the existing limitations determined upon by the transport line are influenced primarily by the parameters of the engine manufacturer. Some of the transport lines with a wealth of cost statistics can tell immediately what their increase in mechanical failures and maintenance costs will be for a given increase in existing speed. With all these data, however, they still do not know what their power output is, for they do not have any power-output guides. These data cover only indica-

tor engine revolutions for average altitudes. The supposition is that the failures and excessive wear occurred at those altitudes which exceeded the maximum existing speed. But there has been no method for keeping to the slow ones who drag down the schedule to, but not past, the limit.

Limitations on maneuvers

The airplane manufacturer's limitations upon the class of his product include a diving speed limit, a pull-out acceleration limit, a landing stress limit, and sometimes a turning speed limit on rough fields. Sometimes the airplane is built for high-speed diving, and

more recently these diving limits have occasionally been removed so that long vertical dives up to terminal velocity have been considered, while not normal, in the class of cruising and "emergency" in the class of single-engine over-load emergencies. Pull-outs from dives, even for military aircraft, have now even reached the stage of entire removal of limitation, although it seems that this stage has been approached so closely that the stress on the human organism reaches its peak just before the stress on the material of the structure. To limit the airplane which cannot be broken up in the air by the most violent stresses at a pilot level or de-

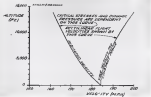


Fig. 1—Variation of time and corresponding indicated velocity with altitude for constant power, constant and dynamic forces, and for an airplane not constrained to dynamic pressure and hence in balanced equilibrium.

striving himself or the pilot completely out of his seat) has been so slow as to cause several people, but the tendency to be paid for such a mistake has not returned them. Neither a performance penalty for an excessively being applied nor a control penalty where the stability and power of the controls are such that they resist the most powerful effort in self destruction, are attractive.

An airplane becomes more and more efficient, and tend more and more also toward careful control of structural weight, its diving and acceleration limitations on conventional transport types approach those used in aerial flight. Thus for a modern transport airplane which can cruise at 200 miles per hour, it might well be considered wise at the altitude to subject it to indicated speeds 25 per cent above that existing speed. It is quite probable that in the near future existing speed, ma-

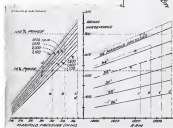


Fig. 2—Variation of available power and engine revolutions with both indicated speed and existing speed. Although these curves are for one level aircraft, all these limits for one aircraft which can be applied to others by the following: (a) Existing 100% W.P. (b) Existing 100% W.P. (c) Existing 100% W.P. (d) Existing 100% W.P. (e) Existing 100% W.P. (f) Existing 100% W.P. (g) Existing 100% W.P. (h) Existing 100% W.P.

be reduced on airplanes which cannot be used fully at indicated speeds is high as their indicated existing speed. This assumed practice will be done upon more in length later. It can be approached more readily when it is recalled that stresses and dynamic forces on an airplane are proportional to dynamic pressure and hence to indicated speed, while cruising speed at constant power increases with altitude, although indicated speeds also, increasing. This relationship is illustrated in the graph of Fig. 1.

The airplane is designed for certain stresses normally realized in normal

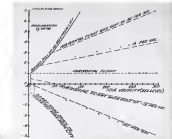


Fig. 3—Variation of available power with both indicated speed and existing speed. Although these curves are for one level aircraft, all these limits for one aircraft which can be applied to others by the following: (a) Existing 100% W.P. (b) Existing 100% W.P. (c) Existing 100% W.P. (d) Existing 100% W.P. (e) Existing 100% W.P. (f) Existing 100% W.P. (g) Existing 100% W.P. (h) Existing 100% W.P.

fast existing and emergency operation for his product in terms of his specifications or guarantees. Creating a normal operation at usual existing speeds. These speeds are those corresponding to 75 per cent power in most instances. He is naturally interested in operating the engine to his airplane at the maximum possible existing power for which the engine builder will guarantee it. For in this condition he will obtain the optimum possible airplane performance and therefore the best sale value. The reports to engineers operating task overloads at diving, pull-out, full-power flight.

Data from the engine builder

The last and most specific deliveries and limitations for existing conditions come from the engine builder whose data on engine stresses in relation to power are more detailed and rich in content for deriving creative limitations than those of any of the other agency. For our power determination we need go to the engine manufacturer whose thousands of hours of testing under carefully controlled conditions serve as the principal guide to the designer lines, the military service, and private firms. By operating engines at stresses over-load until they break down he can determine which parts need attention in the recurrent process of pushing up power output. He has studied this process under of possible combinations.

of conditions to be met in service from the extremely high (up to 4) to the very low (down to 0.5) progressively high supercharging in an emergency takeoff. He has operated engines at rated power for long periods with extremely fast oil temperature effect on wear and tear. He has obtained one thousand hours of engine operation under all sorts of conditions together with records of wear, replacement requirements, cost of operation, and general engine functioning. He knows the effect of a few seconds of overheat (at takeoff from a hot field), of many minutes of high cranking load (for starting up on cranking cylinders), and of the hour after the engine at low power cruising. Knowing this he is in the top position to specify how the engine should be operated in various situations like air maintenance units for continuous cruising power output.

The engine builder is very positive and definite in his statement of operating limits for each part of the transport flight. He specifies the exact loading for engine revolutions, for Mean Effective Pressure, for cylinder temperature, and for fuel consumption. That these limits are quite different for the operation of climbing, of climb, of take-off, and all emergency is to be expected from his demonstration that engine reliability and wear are functions of rate of operation under certain stresses, as well as of the quantitative value of the stresses. These values are given in the following table for two recent engines of approximately 700 hp each. The differences in the values between the two engines are of the order of 10 per cent. The SDG is a higher speed engine than the F3 but of lower displacement and weight (4.5 per cent) less power.

The engine builder is very positive in the fact power of 600 g.p.m. and to the first power of the Mean Effective Pressure. These are the only two quantities that Mean Effective Pressure depends on. It is just as large a factor in power output as engine revolutions. It largely determines the choice of engine wear and bearing loads. Its influence is considered only as important by the engine builder, if not more so, than the engine revolutions. The effect of the combination of manifold pressure and engine revolutions can be best illustrated in Fig. 3 which shows an one sea level, manifold pressure or power for various engine revolutions and on the other side, manifold revolutions to power for various manifold pressures. Full rated power has been indicated by the top horizontal line, and 75 per cent power by the heavy dashed line. It is quite clear from this chart that one may take much power at low manifold pressure and engine revolutions and manifold pressure constant. The limiting revolutions for each operation and the limiting pressure for each operation are shown on the graph by the heavy lines labeled accord-

Engine Manufacturer's Limitations for Peak & Mean Effective Pressures

Engine	Max. Eff. Press. (psi)		Max. Oil Temp.
	Peak	Mean	
Cessna	150	31	400
Clad	150	24	350
Wright	210	33	475
Emergency	250	33	475

Engine Manufacturer's Limitations for Weight Excesses

Engine	Stroke		Max. Oil Temp.
	Peak	Mean	
Cessna	100	5	400
Clad	150	5	350
Wright	200	5	475
Emergency	210	5	475

ingly. Then the area under each set of heavy lines is the available area for the operation in question.

The most noteworthy point to be gleaned from this study of these tables and figures is that during operation, but outside of, up to the manufacturer's limit for cruising engine revolutions comes no greater wear or scrubbing at constant power than during operation up to the manufacturer's limit for cruising manifold pressure. One might look at power partly from the engine revolution standpoint and as to look advance it what seems to be high cruising engine revolutions while keeping nearly constantly high manifold pressure. There is no more justification for the transport use to adjust to cruising at 1,500 r.p.m. if 75 per cent power is needed from the engine to cruise at 32 in. of mercury intake manifold pressure. It is evident that the engine revolutions regarding the engine revolutions limitation so often associated for emergency fans at manifold pressure limitation since the latter is not pertinent to constant steady manifold pressure engine for the worst emergency while the revolutions are

fully allowed another hundred or so. This significant fact is seen in the case of the low speed engine limitations above. The engine builders point to the fact that for constant 75 per cent power, the operation at 1,500 r.p.m. reduces some of the bearing loads induced by explosion pressures and it occurs as the engine then would be a 75 per cent power cruising with regular pitch set up to limit the g.p.m. to 120 with constant manifold pressure for the power of some 34.5 in. of critical altitude. On the airplanes which demonstrate increased velocity diving, the engine revolutions during the dive are allowed to go up to as high as 1,800 r.p.m. but intake manifold pressure is strictly limited by requiring that the throttle be closed during the dive. A 25 per cent aviation over normal manifold pressures would be much better as the engine fans a wider increase in revolutions. This point is still not fully appreciated by many transport operators.

Constant cruising is not simply a power function, much less is it an engine-revolution limitation. It must be an answer to the whole two-stroke, air-coupled, airplane's make. Improved cruising means a more powerful engine in the air, not one that quickly, safely, comfortably, economically in these dimensions.

This article has presented an outline of present methods of cruising control with a definition and delineation of cruising conditions. It has attempted to show that present conditions of cruising are inadequate to meet present demands of operation. The writers plan to discuss in the next issue some of the conditions in which they will define the operational being produced for determining cruising performance and present a few methods of suitably designed engine performance and economically controlling engine operation.

Sodium Beacon

DUTCH glazing engineers have patented a new type of lighting used for runway beacons, which is described as giving perfect lighting qualities. The three sodium

lanterns has been suspended side at the Walkway Airport at Rotterdam.

A 1,800-watt set, which emits 5,000 candlepower consists of six glass lanterns fixed together, each 30 in. long, containing one-filled lamp tubes. A certain amount of metallic sodium has also been introduced into each lamp tube. When the tubes are cold the sodium is deposited in metallic form on the inner surface, but as the tubes warm up the sodium vaporizes and emits its characteristic yellow light. On the basis of former control per percent watt input, these lights are approximately five times more efficient than the average incandescent type.

See reflectors of any sort are used. The tubes, which are 100-watt, are simply mounted on a metal tripod.

EDITORIALS

AVIATION

EDWARD P. WARTER, Editor

Notes on a Great Confusion

THREE MONTHS AGO the average citizen of the United States thought about aviation but little or not at all. He read the air mail occasionally, and knew by his sales that it had been wrong through properly to their destination. He might occasionally travel on the airlines, or if he failed to do that he would have had plenty of friends who did so, and he knew in a general way that it was possible to go from practically any where to practically anywhere else at a very high rate of speed and at a fare not much exceeding that of the railroads, and that accidents were rare enough so that there seemed to be no occasion to worry about them. He had seen news items at the Air Corps performing military exercises and of casual aircraft maneuvering from the aircraft carrier, and he had marveled at the skill displayed and at the perfection of the organization. But he never got very much excited about aviation. It seemed to him perfectly obvious that aviation was making tremendously rapid progress, and that appeared a sufficient evidence that it was on the hands of people who knew how to do the job. It seemed plain on the face of the matter that things were going remarkably well, and he was quite content to let it go as that.

But that was last December or thereabouts, and this is March. The same gentleman now seems eagerly glad to have morning newspaper and looks to the top of page 1 to see what new accidents have been alleged, what new organizations have been envisaged and what new crashes have occurred. On the editorial page of the paper he feels at least once a week, and on some cases nearly every day, a vehement assault either upon the character and the integrity and the competence of the aviation industry or upon the wisdom of the Administration's action in congressional matters and upon the members of its advisers. Everyone has taken sides. Every newspaper, and very rarely every newspaper reader, has divided himself in one camp or another, and no change is too temporary to be made and to be believed. Aviation has become the leading political issue of the day, and our friend the average citizen, all aware or less keenly aware that he knows nothing at present about the subject, is nevertheless flagging himself up with a choice collection of prejudices and antipathies, not

to mention some very remarkable delusions on matters of simple fact. Public opinion is involved in a fog of impenetrable density. The typical "average" man is agitated, nervous, and something ought to be done immediately, but he has no concrete idea what it should be. Neither, it may be said in passing, have a good many of the people upon whom some part of the responsibility for determining conditions on policy rest obviously not.

From a multiplicity of confusion, completely clarifying, calmly bring wisdom. The disclosures of the last six weeks have produced a real sense but that light. Recognizing that and fact, every reader of Aviation has a new responsibility. We are opposed to be the experts in the field, and it is up to us to enlighten our neighbors. There probably is not a single member in this magazine, unless he lives alone in the middle or an vast wilderness, that hasn't been confronted over and over again with demands for "the low-down on this aviation stuff!" The temptation is to dash straight to newspaper to any such situation with arguments and evidence, but the temptation must be resisted. Patiently, calmly, conscientiously, specifically, we must explain the facts. The case of American aviation is far too strong to be disrupted in mere indignation. Each of us, and that goes for every reader of this paper, must become a personal center of reliable information, and we must protect our confidence that someone, ultimately, the truth will prevail.

Most apparent current fact is that an emergency has been created, not only in an airport but in public opinion as well. There is an urgent demand for action, and if action designed to be the foundation of a permanent policy is evolved into an emergency after the odds are heavy that it will prove to have been foolish.

With so much all going on at once, it is impossible to get any sense of coherence except by focusing attention first on one ring of the circus and then on another. Obviously air policy must be considered as a whole and as parts must be brought into some rational relationship to each other, but as a preliminary step ought to look at each part by itself and to see what good sense and a reasonable interpretation of experience show in connection with each one.

I.

Air Transport

THE MAIN ACT in the air transport ring just at present is the legislative preparation for revising the air mail system. We discuss that elsewhere in the current issue in full detail, so we may confine ourselves here to more general observations. The bill most under discussion at the moment has certain recognizable and a number of bad and tragic and positively workable features, but the attitude of the congressmen who have been holding the hearings gives every reason to expect that substantial improvements will be made before the measure is finally passed and becomes law.

There is the sharpest division of opinion, and the sharpest fight is likely to be centered, upon the status of the existing companies, holders of the existing contracts. Quite apart from any consideration of justice, they are threatened with lowering the virtues of a political struggle in which certain of the interests of the Administration are implicitly involved that the companies are at least their major officials must be permanently disbanded from any future air transport operations, and that the disbandment must be accomplished without any resort to court proceedings at a justification of the summary action already taken.

We refuse to believe that any such attitude is at all general. We earnestly hope that the transport operators will maintain themselves firmly and unflinchingly upon the ground of the right of every man, even the head of their employees, to have a trial in accordance with the provisions of law before being subjected to any penalty, and in particular before being forbidden to practice his chosen vocation. We have no sympathy with anyone who may be deliriously proud of any improper act, but we still insist that justice demands that the proof be presented before an impartial tribunal. Eminent conscientiousness have no place in American justice.

Compared with the great issue of the right of trial anything else seems trivial, but there are other matters that deserve mention. Foremost among them is the necessity perhaps not even yet sufficiently emphasized, of expert-witness personnel for an air transport line. Not only must there be a few experienced pilots, but the organization itself be experienced as an organization. It is not too much to say that any group that applies for an air mail contract should be required to show that they have actually been engaged in operating some kind of an airline for a reasonable period and with satisfactory efficiency. The potential effects of a provision that would allow anyone who could raise the money for a bond to put in a bid, and then give five or six months to get together an organization and decide how they were going to operate after the contract had been awarded, would be simply dreadful. Equally dreadful would be the results of a subdivision of such of the transcontinental routes into a number of short stretches, independently allotted to different contractors and inde-

pendently operated, yet it appears that such a scheme has actually been seriously considered.

To offset all these alarms at least one piece of good news, and that is that practically all hands seem to be agreed that the allocation of air mail packages to be freighted separately between the transport operators and the government ought as soon as possible be taken out of the hands of the Post Office Department and put into those of a non-political commission. The readers of AVIATION will remember that we have approved that theory for a long time past and it is with profound relief, in the middle of all the present turmoil, that we see the commission idea apparently on the point of gaining general acceptance.

II.

The Army Flies the Mail

THE ARMY AIR CORPS has suffered severely through the air mail cancellations. It has lost four men actually flying the mail, and seven more in preliminary practice and as ferrying trips and the like. It has had about half a million dollars' worth of equipment written out. Finally, and by a most late account, it has been the victim of an utterly unscrupulous line of public confidence. The accidents that have attended an attempt to take over the mail routes have been handled in showing the total indifference of military aviation, the inferiority of its equipment and its personnel, and the senselessness of its organization. All that is practically chronically pure nonsense, and it deserves to be shown up as such.

To abbreviate a story which could be told in enormous length if space permitted, there were several basic troubles in the way of the Air Corps being able to carry the mail successfully. They were plain, from the first to anyone who knew how commercial aviation was operated, and also knew something about the military service, and there were plenty of people in the non-military field who made grossly accurate judgments of what was going to happen as soon as they heard of the name of the order in the Army.

Possession of the Army's afflictions was the failure to realize the extent to which successful commercial operations in bad weather depends upon the pilot's specialized knowledge of the route that he flies. When the time comes for getting into an emergency field at night and in a snow squall the pilot to get away with it successfully will be the pilot that has circled the field repeatedly, that has studied its every obstruction, and that has made practice landings in it by day under all manner of wind conditions. The newcomer on the route, with only the boundary lights and obstruction lights to serve as his guide, is going to crash up under conditions that would never cause a moment's worry to the man to whom flying that particular route has been a profession

and the study of the route a constant preoccupation for months or years. To the Air Corps officer, however great his skill as a pilot and as a navigator may be, the little group of lights showing up through a mist in the darkness on the route that he has not undertaken to fly a couple of weeks ago is just a group of lights. To the old timer on a commercial line, the same group spells Altoona or Pines Grove or Pines City as plainly as though the name had been printed out in letters of fire.

The second great difference between the Air Corps and the commercial operators, and the second great handicap to the Air Corps in attempting to do a transport job, has been the difference in intensity of the typical pilot's flying experience. The average Air Corps pilot gets 200 hours of flying a year, and the officer that gets more than 400 is a rare bird indeed. The first pilot on an airline pilot is somewhere between 800 and 1,000, and he does it year after year, with an unintermittent tour of departmental duty behind a desk to cut his flying time to a hundred hours a year or less. Limitations on appropriations and the wastefulness of the Budget Bureau over any apparent extravagance with aircraft goods play a big part in the difference, but economy is not by any means the whole story. A more serious factor is that whereas the Air Corps has to be ready for war tomorrow, it also has to be ready for war five years from now. It was never absolutely sure to come within the next 60 days, it would be possible to concentrate on training dispatches in the Air Corps to be good number 3 men in fighting formations to the total exclusion of every other element of their education. But if that were to be our course, and if we were to discontinue it by failing to arrive at a schedule, we would have a lot of spectacularly skilled junior officers who had been doing nothing whatever to prepare themselves for the higher commands into which they would naturally in this course be promoted.

The pre-arranged and in fact the only professional obligation of the airline pilot is to carry loads of passengers and mail sensibly, safely, uneventfully from place to place on regularly scheduled flights. Everything that is any degree commensurate to his ability to do that job, he must study, everything outside its bounds is more or less irrelevant from a narrowly professional point of view. The officer of the Army Air Corps has no such specifically limited list of duties. Parts of all he has to fly, but beyond flying he has to do a wealth of paper work—he has to possess occasional administrative offices that leave him little time for flying even if administrative affairs and related gasoline were available. He has to go from time to time to a variety of specialized schools for training in the art of war—he has to serve very often as supply officer or engineering officer of a squadron—and at a very early stage of his career he begins to have extra duties on his shoulders and to have to take responsibility for their welfare and their training. The airline pilot, in short, even though in some cases he will ultimately become an executive of a line function for

the time being purely as a professional pilot able to concentrate entirely upon flying. The junior officer in the Air Corps, on the other hand, has to be considered not only as the second best person of today but also as the major of the years to come, and he has not only to be trained for his present job as a junior staff in a squadron but prepared also for the day when he will command a group. That problem exists in every military service in the world, and not even an increase of appropriations will change it. The officers of the army can't be made precisely efficient in doing the jobs of the commercial airline without at the same time losing ground in their ability to perform their own function, of which actual personal flying is only a part, in many cases a small part.

Equipment lists of course played a part, and it is difficult for the first time that we accurate issues for real concern and serious inquiry. Surely the Army plane is handicapped as against the specialized transport type. Added weight costs too much in fighting ceiling, and the space left for instruments and the like after the necessary utility equipment has been installed is limited. Nevertheless, making allowance for all these factors, it will seem that the record of the Army's organizational equipment, and most especially of radio, was disappointing if not deplorable. The report of non-functioning radios was discouragingly frequent in connection with forced landings, and service personnel ought to learn attention there for what may be some very salutary lessons. Navigating equipment was not the major cause of the Army's troubles, but it was a contributory one, and it is distinguished from the other causes in that it should normally be subject to some modification without sacrifice of military considerations.

Finally there is airplane design. Of that much might be said, but one point is at once typical and unrepresentative. The old government-operated or mail service hadn't been making the mail for two years, suddenly after the War, before it had learned the necessity of getting the mail in a jet between the wings and getting the pilot well back toward the tail for safety in a crash. The same arrangement had persisted as practically all of the planes now used for mail flights when the weather is too bad to risk going out with passengers. The Army on the other hand necessarily flies the pilot's location from ordinary considerations and puts the mail where the observer would be. With the result that the pilot is in the most hazardous position possible and that a maximum in mail load unless fully compensated for by ballast, has as an immediate and in some cases a really serious effect upon the longitudinal stability of the airplane.

Under all these handicaps and others of lesser order, the wonder was not that the Army had accidents, nor that it failed to equal the privately-operated service in efficiency, but rather that the accidents were no more numerous and that the operating record was so good as it was.

III.

Military Aircraft Procurement

THE PROCUREMENT PROBLEM can be formulated up in one single wording: *disturbance*. If the War or Navy Department should decide, or if Congress should like to enforce, purchase by using general type specifications and seeking competitive bids the inevitable effect will be to lower the efficiency of our air services and to render our air force equipment policy in part about as complicated and disastrous as was in the flying of the air mail got into in the closing days of February. Experience at home and abroad as well as the application of plain common sense in the light of a knowledge of the way in which airplanes are designed and built, all agree in pronouncing disaster shortly thereafter. If either the Congress or the Assistant Secretary of War set out to make a fetish of competitive bidding on military airplanes, they will be playing with plenty dynamite—a fact which will become only too apparent in the course.

Less obviously disastrous is the scheme of putting a substantial part of the construction of service aircraft into government-owned plants, but of that more, too, the effects are almost certain to be bad. It is of course perfectly possible for the Army and Navy to recruit a specialized force of aeronautical engineers who will make a professional specialty, out of aircraft design and take the sole responsibility for the initiation of orders and naval aircraft projects, exactly as the crews of naval construction now controls the design of all warships. That doesn't seem to us the best way of getting around our aerial ideas into military airplanes, but it is at least a possible scheme.

To single government design and construction with design and construction by private firms, however, threatens a sacrifice of the virtues that either system by itself might possess. The service personnel will be restricted to a limited amount of design and production. They will be the inevitable nature of things be developing ideas and enthusiasm of their own which they will want to push, and they will find themselves in the position of being at the most time competitors in a contest and the judges of its results. It is extremely difficult for the best of men to preserve a judicial attitude under those conditions. The most ingenious and the most successful of private constructors is then likely to be the one who attacks very closely to the doctrine that the service personnel have embodied in their own designs, and the restriction of private initiative and the inhibition of the free play of imagination in the development of military airplane designs in the industry are inevitable. These have always been the consequences of government entry into the business of aircraft construction or even into the field of aircraft design on any large scale, whether here or in European states, and there is no reason to suppose that the experience of the future will differ materially from that of the

past. The trend of government policy, and especially the threat of Congress upon procurement matters, seems to be toward the adoption of practices sure to retard progress and to slow down an important and military aircraft performance which has been extraordinarily rapid over the past three years.

IV.

Building an Air Policy

THERE ARE other rings in the circus and other aspects of an aeronautical program, but for the moment at least they mean less important. Some time in the very near future it will become apparent that policy must be considered not only in its several parts but also as a whole. That does not mean that all aeronautical activities need be centered under a common head, and indeed any carrying of military and commercial aviation would be bad for both and disastrous for the latter, but it does call for a certain consideration of the relations of the aeronautical interests in the various departments of the government to each other and of the ways in which they can best be made to supplement each other without losing their efficiency in their own primary functions. To the end of making such a study, the time is more than ripe for the appointment of a Presidential board of distinguished citizens of varied interests and points of view who can consider this as a public service of immeasurable importance and who can plan to give to it at least the major part of their time for a number of months. Certainly there is no field in which the value of such an inquiry would be greater.

It is more than eight years since there has been a general examination of the air policy of the United States, and during that time aircraft have appeared and the masses of aerial operations have changed out of all recognition. Now, to emerge from victims of the wilds of fanaticism with a will to serve or of progressivism with something to sell, we should survey the field once more. The President cannot possibly take the time to do it himself. Neither can executive officials of his Administration, who would in any case find it difficult to command the detachment that a broad survey of the whole scope and prospect of air policy demands. Those whose concern is not with the promotion of any particular corporate or individual interest, but with the good of American aviation as a whole and with its continued development of steadily improving service to the American people and steadily increased strength as an arm of their defense, will breathe easily for the first time in many weeks when they hear the announcement of a Presidential determination to appoint a special board to inquire. From then to start, our American aviation and to the best and most successful ways of promoting its progress along the lines of maximum utility.

NEWS OF THE MONTH

Mad aeronautics

TEN FATALITIES among Navy aviators occurred in 1938 over the mail was issued and a growing role of public attention against the possibility of using the Air Corps as a casualty list when they had obviously not been trained, elected as order from President Roosevelt through Secretary of War Dorn on March 30 in "any aircraft of mad expert water such conditions as would insure against constant recurrence of fatal accidents."

The following day Maj Gen B. D. Foshell, Chief of the Air Corps, announced that mail operations would be suspended entirely until a higher safety factor could be established. At the same time he ordered that no pilot be acceptable in the development of military flying, that the emergency had been called by him as an actual procedure test for the Army, and that furthermore the percentage of mortality had been higher than any other in any war was mentioned including another and other conditions. He then directed an inspection tour over the mail routes of the Eastern and Central Zones.

It was in a that the greatest possible number of military mail pilots, it is reported, should be sent to throughout the service. In carrying out his orders plans and personnel were redistributed, better mail and night-flight operations were established, and mail planes, which carry only a small mail load, were discarded in favor of bigger ships. Now that Martin bombers are in order for the Army, will be added to the mail service as soon as delivery is made. Army aircraft and the Post Office Department, together with the National Aeronautics Administration, selected on the basis of the percentage formerly served as well as on the basis of safety.

Meanwhile another investigation committee visited the Air when Secretary Dorn instructed General Dorn, Foshell, Stevens, Gault, and Kilbourne when Foshell is the only one with actual experience in aviation to look into the matter of Air Corps efficiency in general, and specifically into its mail operations. Three of the most highly published names in the aeronautical world were added as non-military members of the committee: Maj Gen Wright, chief of the Air Corps, who was the only one of the two who felt disposed to accept the honor. Gen Wright, presently at Kille Black 30 years ago, was obliged to decline be-

cause of ill health. Col. Charles A. Lindbergh refused to serve on the grounds that in his opinion the use of the mail service in any mail was similar both to Army personnel engaged in conventional flying and to the transport company, whose contracts had been canceled without hearing. Urged to reconsider by the Secretary, Gen. Calkins Lindbergh telegraphed a second refusal saying that he would not take part "directly or indirectly in the operation of military Service of American business and commerce."

Grateful for the week's respite from abnormally frequent flying duty, Army personnel enlisted in mail service promised, however, that they were withdrawing from both in holding and were cutting back their flight schedules so that Congress had not yet passed the Air Mail Bill providing that with the 35 per cent allowance for extra compensation. Granted by the House (Aviation, March, page 57) the bill was amended by the Senate and returned to the House for approval. The President signed it on March 23.

Private additional benefits for Regular and Reserve personnel killed or injured in mail service, one annual month's pay for each military, if it is reported out now to regular discharge, but also to Veterans' Bureau compensation, based on the war

notes for disability. In case of death the dependents of either Regulars or Reserves would receive the statutory gratuity as well as pay for those widows compensation of \$50 per month instead of the \$32 provided in case of peace. This amendment was made retroactive as well as apply to the next dead flier. Another amendment provides that pilots should not be re-mustered for mail service unless fully equipped for safe day and night mail transport, nor should pilots be assigned to mail duty unless fully trained in the use of such equipment.

Business headquarters of the Eastern Zone were transferred for the third time (Aviation, March, page 52) from Floyd Bennett Field in Brooklyn, New York, to the new Long Beach Field at Garden City, Long Island, because of inability to reach an understanding of the terms of the day-to-day lease offered by Mayor LaGuardia.

Two days before the resumption of service, the seventh Army liability occurred. Reserve Officer H. G. Baldwin, formerly captain of the United Air Lines, crashed on a test flight over Chesapeake Bay.

Though criticized letters were presented carried by the railroad during the winter, it was later decided that large parts of the air mail system had wrapped out of the situation, highly increased to business hours long as continued to the world service furnished by air, by wrapping in a package between to a minimum duration and making them as air express. Nothing in the Post Office regulations forbids this practice, provided that the bill or mail is placed in a special container, and given a special label. This has been done in companies disallowed from mail carriage continued in by letters bound in packages which require a \$4.00 rate of some companies at a rate of 125 per cent in express loads. Between New York and Chicago the rate for air express varied between minimum charges of \$1.25 a package to 30 cents a pound to sums of more than 20 lb and commensurately as other lines in proportion to the distance covered.

One week after the grounding order the Army set out again over eight routes estimated to short 30 per cent of those authorized by commercial lines, and about 60 per cent of those first flown by the Army. Of this new mail service the Post Office Department worked at length, and is still advising on technical matters pertaining to rates

Calendar

- April 11—Lectures: Earl K. Long, Public Affairs, Chicago; Franklin D. Roosevelt, New York.
- April 12—Lectures: Earl K. Long, Public Affairs, Chicago; Franklin D. Roosevelt, New York.
- April 13—Lectures: Earl K. Long, Public Affairs, Chicago; Franklin D. Roosevelt, New York.
- April 14—Lectures: Earl K. Long, Public Affairs, Chicago; Franklin D. Roosevelt, New York.
- April 15—Lectures: Earl K. Long, Public Affairs, Chicago; Franklin D. Roosevelt, New York.
- April 16—Lectures: Earl K. Long, Public Affairs, Chicago; Franklin D. Roosevelt, New York.
- April 17—Lectures: Earl K. Long, Public Affairs, Chicago; Franklin D. Roosevelt, New York.
- April 18—Lectures: Earl K. Long, Public Affairs, Chicago; Franklin D. Roosevelt, New York.
- April 19—Lectures: Earl K. Long, Public Affairs, Chicago; Franklin D. Roosevelt, New York.
- April 20—Lectures: Earl K. Long, Public Affairs, Chicago; Franklin D. Roosevelt, New York.
- April 21—Lectures: Earl K. Long, Public Affairs, Chicago; Franklin D. Roosevelt, New York.
- April 22—Lectures: Earl K. Long, Public Affairs, Chicago; Franklin D. Roosevelt, New York.
- April 23—Lectures: Earl K. Long, Public Affairs, Chicago; Franklin D. Roosevelt, New York.
- April 24—Lectures: Earl K. Long, Public Affairs, Chicago; Franklin D. Roosevelt, New York.
- April 25—Lectures: Earl K. Long, Public Affairs, Chicago; Franklin D. Roosevelt, New York.
- April 26—Lectures: Earl K. Long, Public Affairs, Chicago; Franklin D. Roosevelt, New York.
- April 27—Lectures: Earl K. Long, Public Affairs, Chicago; Franklin D. Roosevelt, New York.
- April 28—Lectures: Earl K. Long, Public Affairs, Chicago; Franklin D. Roosevelt, New York.
- April 29—Lectures: Earl K. Long, Public Affairs, Chicago; Franklin D. Roosevelt, New York.
- April 30—Lectures: Earl K. Long, Public Affairs, Chicago; Franklin D. Roosevelt, New York.

American airplane specifications

Assumes that we assume responsibility for the figures given.

Manufacturer	General		Power Plant		Weights		Performance	
	Design No.	Year of Invention	Engine (Type, No., HP)	Max. Altitude (ft.)	Empty Weight (lb.)	Max. Gross Weight (lb.)	Max. Speed (mph)	Max. Range (miles)
SINGLE ENGINED PLANES WITH SEATS FOR FEWER THAN FOUR PERSONS								
Boeing Stearman	100	1925	Boeing Model 100, 100 HP	10,000	1,200	2,000	110	1,000
Cessna 170	170	1928	Cessna 170, 100 HP	10,000	1,200	2,000	110	1,000
Grumman PT-17	PT-17	1925	Grumman Model 17, 100 HP	10,000	1,200	2,000	110	1,000
Stearman	100	1925	Boeing Model 100, 100 HP	10,000	1,200	2,000	110	1,000
SINGLE ENGINED PLANES WITH SEATS FOR MORE THAN THREE PERSONS								
Boeing Stearman	100	1925	Boeing Model 100, 100 HP	10,000	1,200	2,000	110	1,000
Cessna 170	170	1928	Cessna 170, 100 HP	10,000	1,200	2,000	110	1,000
Grumman PT-17	PT-17	1925	Grumman Model 17, 100 HP	10,000	1,200	2,000	110	1,000
Stearman	100	1925	Boeing Model 100, 100 HP	10,000	1,200	2,000	110	1,000
MULTI-ENGINED TRANSPORT PLANES								
Boeing Stearman	100	1925	Boeing Model 100, 100 HP	10,000	1,200	2,000	110	1,000
Cessna 170	170	1928	Cessna 170, 100 HP	10,000	1,200	2,000	110	1,000
Grumman PT-17	PT-17	1925	Grumman Model 17, 100 HP	10,000	1,200	2,000	110	1,000
Stearman	100	1925	Boeing Model 100, 100 HP	10,000	1,200	2,000	110	1,000
AIRBUSONS AND FLYING BOATS								
Boeing Stearman	100	1925	Boeing Model 100, 100 HP	10,000	1,200	2,000	110	1,000
Cessna 170	170	1928	Cessna 170, 100 HP	10,000	1,200	2,000	110	1,000
Grumman PT-17	PT-17	1925	Grumman Model 17, 100 HP	10,000	1,200	2,000	110	1,000
Stearman	100	1925	Boeing Model 100, 100 HP	10,000	1,200	2,000	110	1,000
AUTOGYRO								
Boeing Stearman	100	1925	Boeing Model 100, 100 HP	10,000	1,200	2,000	110	1,000
Cessna 170	170	1928	Cessna 170, 100 HP	10,000	1,200	2,000	110	1,000
Grumman PT-17	PT-17	1925	Grumman Model 17, 100 HP	10,000	1,200	2,000	110	1,000
Stearman	100	1925	Boeing Model 100, 100 HP	10,000	1,200	2,000	110	1,000

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American airplane specifications (continued)

Approved data and accuracy responsibility for the figures given

Manufacturer and Description	A.T.C. Number	Approved TC		Area (sq. ft.)		Wings		Span		Length		Height		Weights		Performance		Propulsion		Other		
		Max. Gross	Max. Ramp	Span	Wing	Span	Wing	Span	Wing	Span	Wing	Span	Wing	Span	Wing	Span	Wing	Span	Wing	Span	Wing	Span
SINGLE ENGINE PLANES WITH SEATS FOR MORE THAN FOUR PERSONS																						
Boeing Stearman		11000	11000	35	110	35	110	35	110	35	110	35	110	11000	11000	110	110	110	110	110	110	110

SINGLE ENGINE PLANES WITH SEATS FOR MORE THAN THREE PERSONS

Boeing Stearman	11000	11000	35	110	35	110	35	110	35	110	35	110	11000	11000	110	110	110	110	110	110	110
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MULTI-ENGINE TRANSPORT PLANES

Boeing 247	11000	11000	35	110	35	110	35	110	35	110	35	110	11000	11000	110	110	110	110	110	110	110
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AMPHIBIONS AND FLYING BOATS

Boeing 247	11000	11000	35	110	35	110	35	110	35	110	35	110	11000	11000	110	110	110	110	110	110	110
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American engine specifications

Approved data and accuracy responsibility for the figures given

Manufacturer and Description	Model	Horsepower		Torque		RPM		Stroke		Compression		Fuel		Ignition		Cooling		Other	
		Max. Gross	Max. Ramp	Max. Gross	Max. Ramp	Max. Gross	Max. Ramp	Max. Gross	Max. Ramp	Max. Gross	Max. Ramp	Max. Gross	Max. Ramp	Max. Gross	Max. Ramp	Max. Gross	Max. Ramp	Max. Gross	Max. Ramp
Continental	4-70	70	70	70	70	2400	2400	4	4	16:1	16:1	85	85	12	12	Water	Water	Water	Water

Model	Year	Power	Weight	Dimensions	Other
Model 1	1931	100	1000	100x100x100	
Model 2	1932	150	1500	150x150x150	
Model 3	1933	200	2000	200x200x200	
Model 4	1934	250	2500	250x250x250	
Model 5	1935	300	3000	300x300x300	
Model 6	1936	350	3500	350x350x350	
Model 7	1937	400	4000	400x400x400	
Model 8	1938	450	4500	450x450x450	
Model 9	1939	500	5000	500x500x500	
Model 10	1940	550	5500	550x550x550	
Model 11	1941	600	6000	600x600x600	
Model 12	1942	650	6500	650x650x650	
Model 13	1943	700	7000	700x700x700	
Model 14	1944	750	7500	750x750x750	
Model 15	1945	800	8000	800x800x800	
Model 16	1946	850	8500	850x850x850	
Model 17	1947	900	9000	900x900x900	
Model 18	1948	950	9500	950x950x950	
Model 19	1949	1000	10000	1000x1000x1000	
Model 20	1950	1050	10500	1050x1050x1050	
Model 21	1951	1100	11000	1100x1100x1100	
Model 22	1952	1150	11500	1150x1150x1150	
Model 23	1953	1200	12000	1200x1200x1200	
Model 24	1954	1250	12500	1250x1250x1250	
Model 25	1955	1300	13000	1300x1300x1300	
Model 26	1956	1350	13500	1350x1350x1350	
Model 27	1957	1400	14000	1400x1400x1400	
Model 28	1958	1450	14500	1450x1450x1450	
Model 29	1959	1500	15000	1500x1500x1500	
Model 30	1960	1550	15500	1550x1550x1550	
Model 31	1961	1600	16000	1600x1600x1600	
Model 32	1962	1650	16500	1650x1650x1650	
Model 33	1963	1700	17000	1700x1700x1700	
Model 34	1964	1750	17500	1750x1750x1750	
Model 35	1965	1800	18000	1800x1800x1800	
Model 36	1966	1850	18500	1850x1850x1850	
Model 37	1967	1900	19000	1900x1900x1900	
Model 38	1968	1950	19500	1950x1950x1950	
Model 39	1969	2000	20000	2000x2000x2000	
Model 40	1970	2050	20500	2050x2050x2050	
Model 41	1971	2100	21000	2100x2100x2100	
Model 42	1972	2150	21500	2150x2150x2150	
Model 43	1973	2200	22000	2200x2200x2200	
Model 44	1974	2250	22500	2250x2250x2250	
Model 45	1975	2300	23000	2300x2300x2300	
Model 46	1976	2350	23500	2350x2350x2350	
Model 47	1977	2400	24000	2400x2400x2400	
Model 48	1978	2450	24500	2450x2450x2450	
Model 49	1979	2500	25000	2500x2500x2500	
Model 50	1980	2550	25500	2550x2550x2550	
Model 51	1981	2600	26000	2600x2600x2600	
Model 52	1982	2650	26500	2650x2650x2650	
Model 53	1983	2700	27000	2700x2700x2700	
Model 54	1984	2750	27500	2750x2750x2750	
Model 55	1985	2800	28000	2800x2800x2800	
Model 56	1986	2850	28500	2850x2850x2850	
Model 57	1987	2900	29000	2900x2900x2900	
Model 58	1988	2950	29500	2950x2950x2950	
Model 59	1989	3000	30000	3000x3000x3000	
Model 60	1990	3050	30500	3050x3050x3050	
Model 61	1991	3100	31000	3100x3100x3100	
Model 62	1992	3150	31500	3150x3150x3150	
Model 63	1993	3200	32000	3200x3200x3200	
Model 64	1994	3250	32500	3250x3250x3250	
Model 65	1995	3300	33000	3300x3300x3300	
Model 66	1996	3350	33500	3350x3350x3350	
Model 67	1997	3400	34000	3400x3400x3400	
Model 68	1998	3450	34500	3450x3450x3450	
Model 69	1999	3500	35000	3500x3500x3500	
Model 70	2000	3550	35500	3550x3550x3550	
Model 71	2001	3600	36000	3600x3600x3600	
Model 72	2002	3650	36500	3650x3650x3650	
Model 73	2003	3700	37000	3700x3700x3700	
Model 74	2004	3750	37500	3750x3750x3750	
Model 75	2005	3800	38000	3800x3800x3800	
Model 76	2006	3850	38500	3850x3850x3850	
Model 77	2007	3900	39000	3900x3900x3900	
Model 78	2008	3950	39500	3950x3950x3950	
Model 79	2009	4000	40000	4000x4000x4000	
Model 80	2010	4050	40500	4050x4050x4050	
Model 81	2011	4100	41000	4100x4100x4100	
Model 82	2012	4150	41500	4150x4150x4150	
Model 83	2013	4200	42000	4200x4200x4200	
Model 84	2014	4250	42500	4250x4250x4250	
Model 85	2015	4300	43000	4300x4300x4300	
Model 86	2016	4350	43500	4350x4350x4350	
Model 87	2017	4400	44000	4400x4400x4400	
Model 88	2018	4450	44500	4450x4450x4450	
Model 89	2019	4500	45000	4500x4500x4500	
Model 90	2020	4550	45500	4550x4550x4550	
Model 91	2021	4600	46000	4600x4600x4600	
Model 92	2022	4650	46500	4650x4650x4650	
Model 93	2023	4700	47000	4700x4700x4700	
Model 94	2024	4750	47500	4750x4750x4750	
Model 95	2025	4800	48000	4800x4800x4800	

FLYING EQUIPMENT

**Lockheed Electra
Makes Its Appearance**

IN ANSWER for June, 1931, the design of the Lockheed Electra, first in the preliminary production stages, was discussed in some detail. During the intervening months the machine has been brought into being through the very best processes of manufacture and test-terminating at the flight trials only in March.

Although the Electra in the first instance has had features of the original design as far as several distinctive types of structure, etc. are concerned several details have been altered. Most obvious is the rearrangement of the tail surfaces. The original design contemplated a conventional horizontal stabilizer-vertical fin combination, with the fin built as an integral part of the fuselage in the usual manner. In the ship as produced, the vertical tail struts have been divided into two smaller fin and rudder semi-sections located on the top of the stabilizer. Both stabilizer and vertical fins are entirely self-supporting and free from external bracing members of any kind. The construction can be seen clearly in accompanying drawing and photograph.

The other changes are of a more minor nature. The multiple exhausts which originally had increased wing washability has been revamped as self-cleaning. Conventional pitch propellers have also been fitted—a feature was considered in all new transport airplanes, although



Close-up of Electra's landing gear. Note how multiple exhaust built up and down is secured through the use of a main and gear combination. The main gear struts are visible, with stabilizer head member.



Main reason for the Lockheed Electra's high speed was apparent from this photograph. Note also the semi-sectioned tail construction, an arrangement designed not only for structural construction but to improve maneuverability by placing the end surface in the preferred slip stream.

still in the early stages of application when the Electra was first laid down.

The single open wing is its first feature but a number of interesting features in backbone—a deep flat girder located at about a chord length from the leading edge. This was manufactured out from the side box and all. Forward and shear stresses are taken by the ribs assisted by a broad reinforced member trussing under the smooth covering of the upper surface. The semi-sectioned upper ribs taper according to the general planform of the wing. It appears to be approximately a chord with the wing and about a half chord in width. Subsequent longitudinal struts fastened to each rib serve to stiffen the upper surface. The single center construction is used in the center section as well as in the tip panels. Here however a two bay girder with two parallel ribs members is used. It serves as support for landing gear and engine nacelles, as well as a means of bracing the wing loads to the fuselage. The beam passes through the passenger cabin just aft of the two front seats.

A simple and straightforward arrangement has been selected for the retracting landing gear. The shock absorber, placed at the upper ends to the gear, carry welded steel tube links for Goodrich A-cushions. Segments of boxed wheel axles, built as electrically driven units. Details appear as an accompanying picture.

Space is provided for two passengers, pilot and co-pilot and for all the



insure that go into the making of a modern transport airplane for safety, maneuverability and comfort. Considering its carrying capacity, the Electra is usually compact, a characteristic which accounts for the 225 m.p.h. top speed reported at the flight trial from the total of 810 hp delivered by the two Pratt & Whitney Wasp Junior engines. Cruise speed is said to be in excess of 200 m.p.h.

When test-run completed the first of the Electra's is scheduled for delivery to Washburn Airways for use on high speed schedules between the Twin Cities and Chicago. Pan American Airways also has several orders of this type on order.

THE BUYERS' LOG BOOK

AVIATION'S CARD INDEX OF NEW EQUIPMENTS

This department is equipped to help readers locate manufacturers of any parts, accessories or materials.

AIRPLANE ACCESSORIES

Tire rods (catalog)

The Kelsey Company
Knoxville, Tenn.

CATALOG A-5 (February, 1934) covering all types of aircraft tire-rods available on request. Features multiple road space, streamline (aerodynamic), and tire streamlining rods in all sizes up to and including 18 in. diameter. Can be furnished in half-inch curves, oval, or in stainless steel. Catalog also contains complete engineering data on tire-rods, fittings and accessories.

AVIATION, April, 1934

AIRPLANE ACCESSORIES

Tires

Pirelli Tire and Rubber Company,
Albino, Ohio

A NEW TYPE commercial non-skid road design which is available in the popular 15.00-36 and other sizes has been developed to give high braking efficiency and long wear. Weights are below the range specified by the U. S. Army Air Corps. Made by the Pirelli's grain dipped process. Developed and tested in actual service on the runway-tested Ford transport airplanes.

AVIATION, April, 1934

ENGINE ACCESSORIES

Push rod guides

Air Transport Equipment, Inc.,
Knoxville Field, London City, E. T., N. Y.

PUSH ROD GASKETS of a new type made up of a specially developed non-metallic rubber compound known as Noidal are available for all types and sizes of aircraft engines. These gaskets have been tested under a wide range of conditions by a leading aviation oil company and have been found to be satisfactory. Push rods available in application to manufacturers.

AVIATION, April, 1934

ENGINE ACCESSORIES

Fuel pump

Essex Apparatus Company,
Detroit, Mich.

A NEW TYPE fuel pump, the E-8, is being marketed which incorporates many new features. A flexible valve mechanism eliminates leakage; a spring type seal prevents leakage after long use and an internal relief valve is built in as an integral part of the pump to regulate pressure in the carburetor. The pump will deliver 200-250 lb. per hour flow rate. It weighs 1 1/2 lb.

AVIATION, April, 1934

INSTRUMENTS

Indirect lighting device

Kalman Instrument Company,
1 James Street, Brooklyn, N. Y.

IMPROVEMENT in instrument lighting and mounting mechanism. Individual lamps (E115 lamp) easily replaceable from front of board. No glare or radio interference; intensity controllable. Applicable to planes having no battery system. New, improved mounting permits quick installation and removal from front of board by turning one screw. Flush mounting; sealed apparatus.

AVIATION, April, 1934

PARTS

Engine valves

The Atom Manufacturing Company,
Cleveland, Ohio

THE Wilson-Risk Division of the company has developed and marketed self-cooled valves for airplane and automobile engines. Hollow stems and heads are filled with metallic sodium for cooling. Self-lubrication to insure freedom these valves already have recently been taken by Self-Flow in England and Hinson-Sears in France through the Brook Engine Company, Europe's licensee.

AVIATION, April, 1934

LABORATORY EQUIPMENT

Vibration tester

The Stryer Gyroscopic Company,
Stryer Building, Brooklyn, N. Y.

SPECIAL instrument has been developed for testing vibrating characteristics of airplane instruments or its instrument boards. Consists of heavy base on which a precision indicator pipe graduated in thousandths of an inch is mounted. By pressing the shaft of the indicator against a vibrating body amplification as little as 1/1000 in. may be read directly from the dial.

AVIATION, April, 1934

PARTS

Focusing Devices (catalog)

Fisher-Kelco Corporation,
200 Park Street, New York, N. Y.

THE fourth edition of the Fisher-Kelco Data Book has just been issued and is available upon request. It contains specifications and technical data on various types of self-tapping screws, nails, etc. Although these fasteners are adaptable to a wide range of materials and uses, certain styles are particularly useful in aircraft construction for their small dimensions, etc.

AVIATION, April, 1934

BENDIX

Streamline Wheels and Brakes

These wheels were developed to meet the conditions imposed by tests of a new design. This condition involved design to meet a predetermined form with a maximum maximum of strength and minimum weight within that form. Brake developers, though incidental, was along similar lines, in that maximum controllable power with minimum weight was sought. These structures are a valuable addition to the reduction of parasite in fixed landing gear.

Streamline Tail Wheel
Knuckle Assembly

A series, divided into two parts, designed and built to meet Air Corps tentative standards; 8 and 10 1/2 inch sizes steerable, as well as swivetable 360 degrees for ground handling. Larger size swivetable 360 degrees, but not steerable. All sizes damped to prevent shimmy. Developed by the Air Corps in the quest for reduced parasite and used with streamline tail wheels and tires.



Bendix Pneumatic Struts

The result of more than three years of careful research. Designed in each case for the particular structure into which it is to be fitted. A development carefully worked out to give maximum shock dissipation for minimum load on the structure, and maximum comfort in landing with minimum weight. Can be designed to operate on any size aircraft, both for landing gear and tail wheel work.

Bendix Standard Pilot Seat

Developed to meet both U. S. Army Air Corps and U. S. Navy, Bureau of Aeronautics, requirements. Designed to reduce service stocks to a minimum by the use of a single seat. Utilizes latest knowledge regarding forming of aluminum alloys, the use of electric welding and of special machinery. This seat has passed all tests and weighs, completely painted, six and one half pounds, a major reduction in weight.

BENDIX PRODUCTS CORPORATION
AIRPLANE WHEEL and BRAKE DIVISION

SOUTH BEND

(Division of Fisher-Kelco Corporation)

INDIANA

Round the World Anniversary

TEN YEARS OF DOUGLAS PROGRESS



TEN years after those pioneering Douglas planes established their premature record of being the first to fly around the world, Douglas has introduced its new Transport. ¶ And here again Douglas has pioneered an entirely new standard of air transport luxury and speed. ¶ The three Douglas planes in 1928 averaged 90 m.p.h. for their flying time around the world—a remarkable record then. But on the night of February 18, last, the Douglas Transport averaged 211

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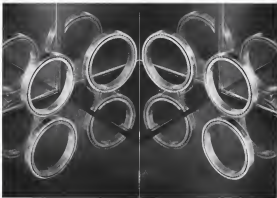
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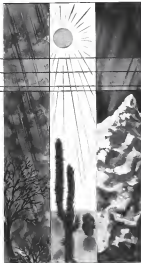
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